

Acknowledgment: This Work Builds on Several Previous NASA Programs

1,000 W/kg Concentrator Study and Roadmap for SSP

Selected for Funding Under the NASA-NSF-EPRI Joint Investigation of Enabling Technologies for SSP (JIETSSP) Program Announcement NSF-02-098

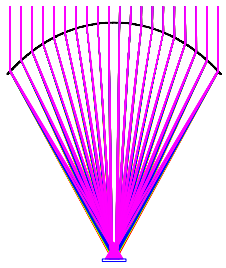
Presented at the Space Solar Power Concept & Technology Maturation (SCTM) Program Technical Interchange Meeting (TIM)

Ohio Aerospace Institute
September 10-12, 2002

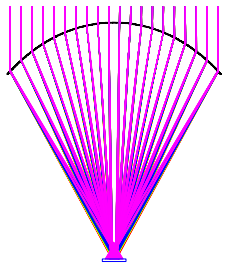
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SEPTEMBER 11, 2001



Program Team and Plan



Study Team

◆ Industry

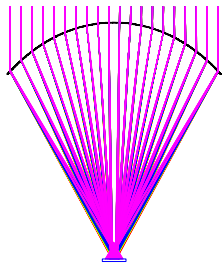
- ENTECH: Prime Contractor and Study Team Leader

◆ NASA

- NASA Glenn: Study Support and Educational Outreach Coordination

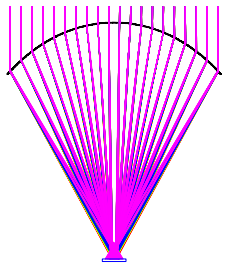
◆ Educational Outreach Partner

- Assumption School – Cleveland, Ohio



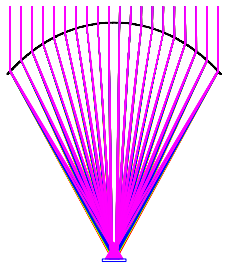
Program Objectives

- ◆ To generate candidate designs of advanced photovoltaic concentrators which are capable of meeting the performance requirements for SSP, including the specific power target of 1,000 W/kg.
- ◆ To perform optical, thermal, and mass analyses of the candidate designs to identify the necessary thicknesses for the optics, radiators, and cell assemblies, to verify adequate performance of these items, and to estimate the areal mass densities (kg/sq.m.) for the candidate designs.
- ◆ To generate a roadmap of the necessary concentrator development program to move from the conceptual designs of this program to prototype components to functional ground-test panels to flight-qualified arrays.



Technical Approach

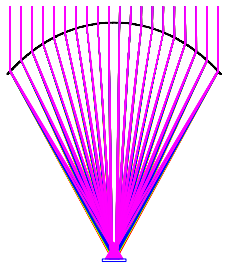
- ◆ For optical studies, use ray trace programs, which include the effects of solar disk size, spectral content of the solar irradiance, optical properties of the concentrator material, and quantum efficiency curves of the multi-junction solar cells. These programs have been used successfully to analyze and design many previous concentrator systems, including the mini-dome lens array on PASP+ and the SCARLET array on Deep Space 1.
- ◆ For thermal studies, use combined conduction/radiation thermal analysis models, which calculate the temperature distribution across the radiator which dissipates waste heat from the solar cell to deep space. The accuracy of these conduction/radiation models has been verified by the measured temperatures on the SCARLET array, which matched pre-flight predictions.
- ◆ For mass studies, use mass estimation spreadsheets, which include the areas, thicknesses, and densities of the components making up the concentrator array, including lenses, radiators, and photovoltaic cell assemblies.



Program Tasks and Schedule

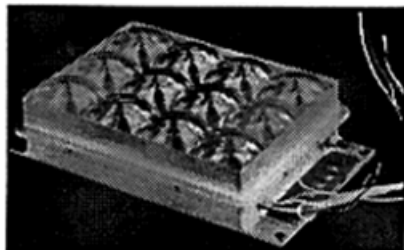
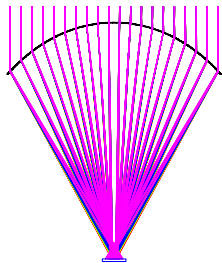
Months After Go-Ahead

Task	Description	1	2	3	4	5	6	7	8	9	10	11	12
1	Generate Conceptual Designs	■	■	■									
2	Perform Parametric Analysis				■	■	■	■					
3	Concept Down-selection and Refinement						■	■	■	■			
4	Roadmapping for Future SSP Concentrator Arrays									■	■	■	■
5	Reviews and Reports	■	■	■	■	■	■	■	■	■	■	■	■
6	Educational Outreach Component	■	■	■	■	■	■	■	■	■	■	■	■



Background and Heritage

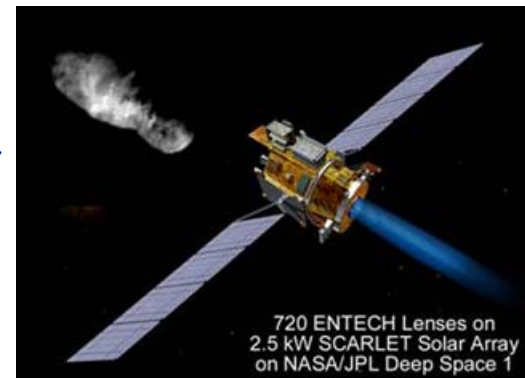
Heritage: Space Fresnel Lens Multi-Junction-Cell Photovoltaic Concentrators



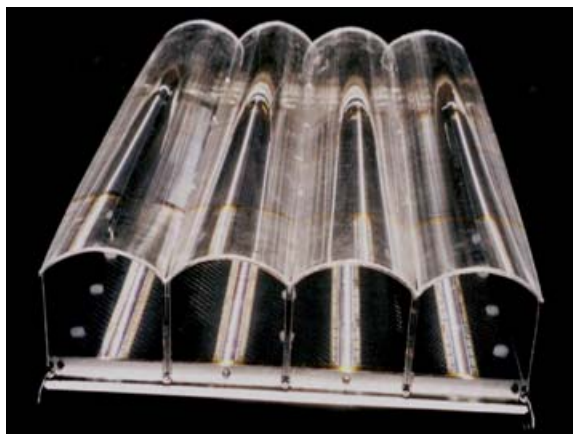
Mini-Dome Lenses
on PASP+ in 1994



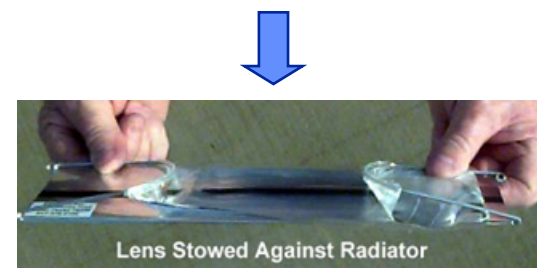
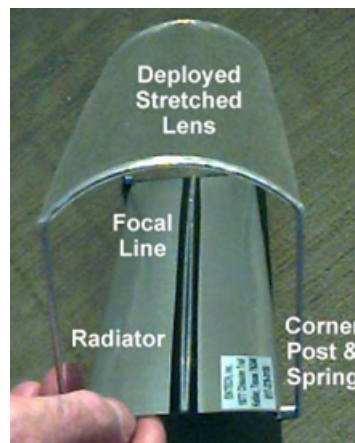
SCARLET 1 Lenses on
COMET/METEOR in 1995



SCARLET 2 Lenses on
Deep Space 1 in 1998

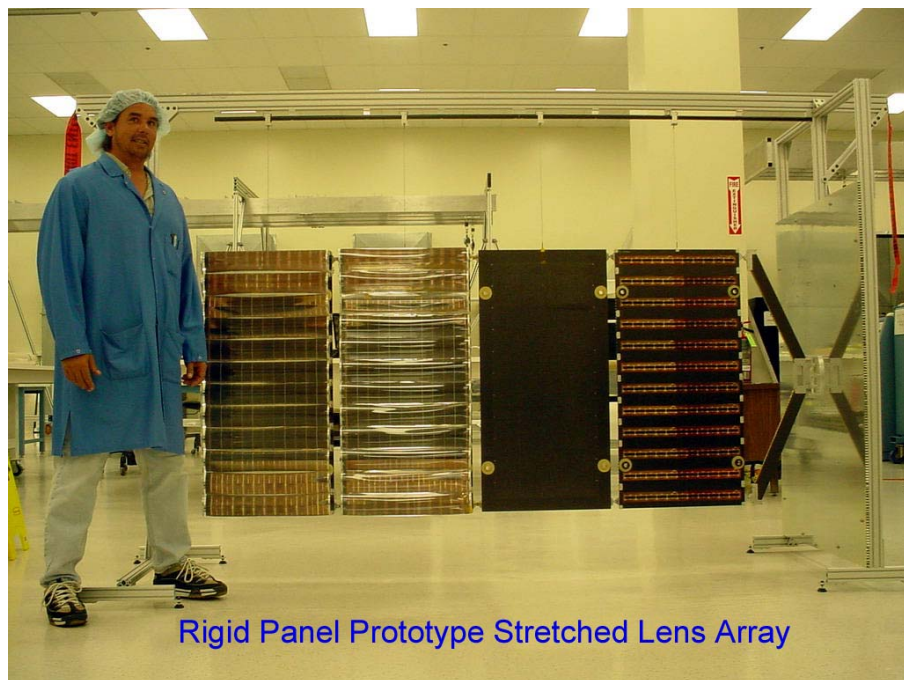
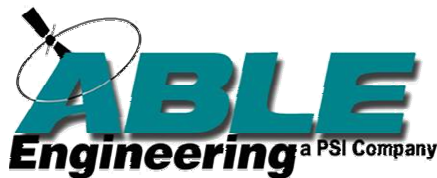


Stretched Lenses and
Photovoltaic Receivers in 2000:
27% Net Module Efficiency

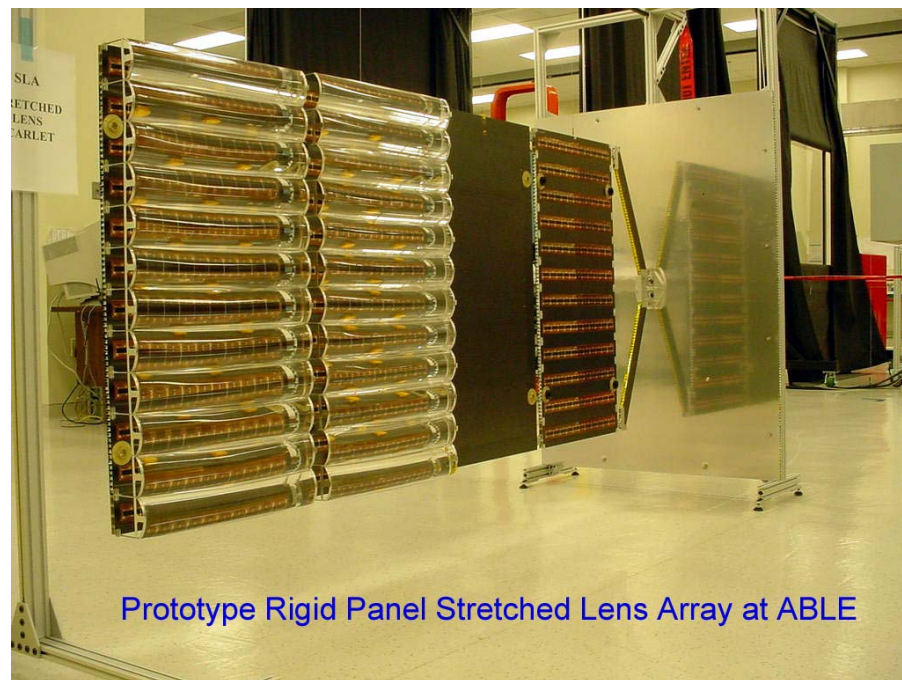


Stretched Lens in 1998

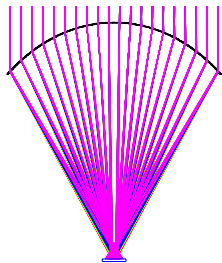
Rigid Panel SLA Prototype Array at ABLE



Rigid Panel Prototype Stretched Lens Array

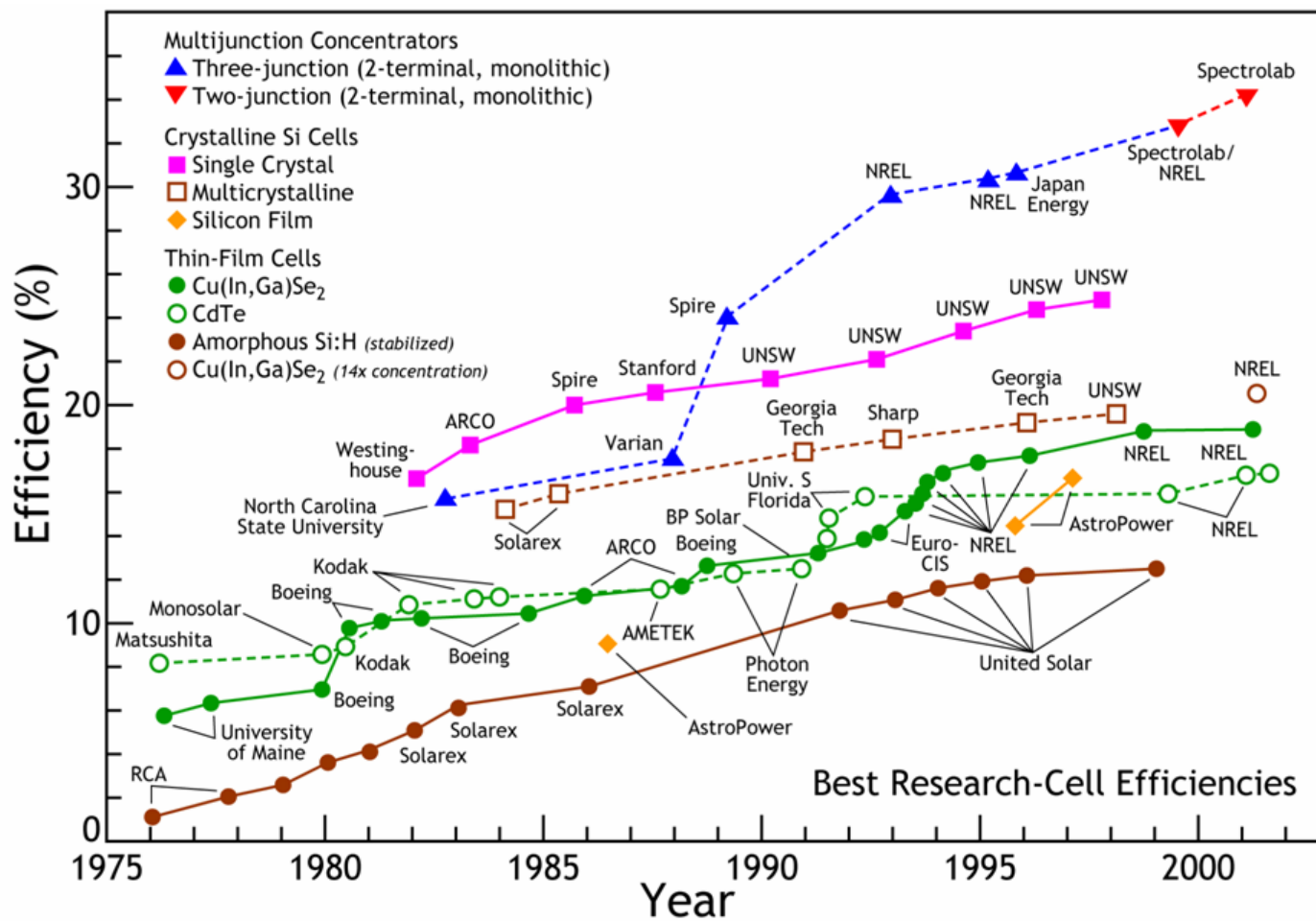


Prototype Rigid Panel Stretched Lens Array at ABLE



Rationale

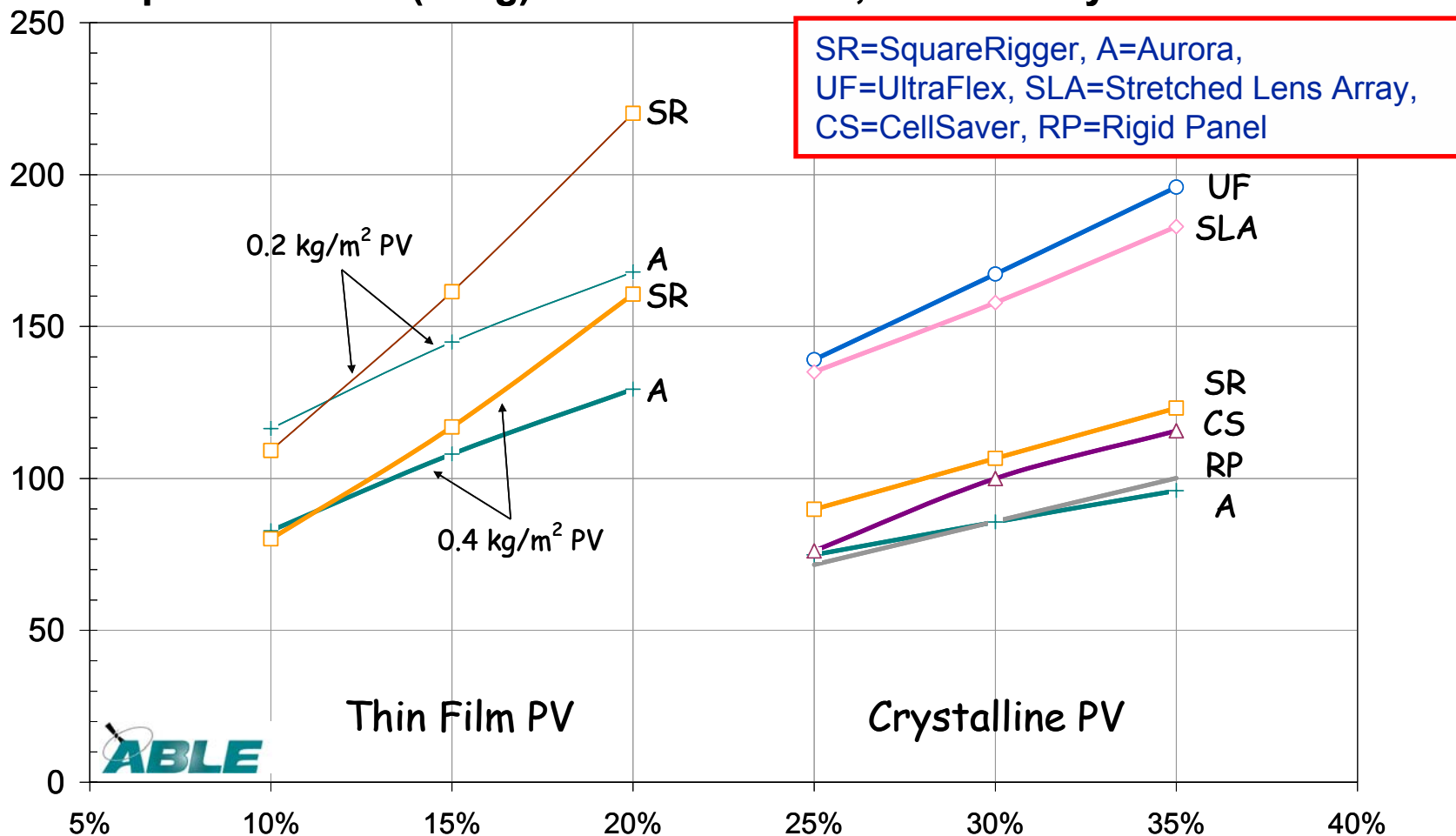
Terrestrial Cell Efficiencies Over Time – from DOE's Strategic Program Review March 2002



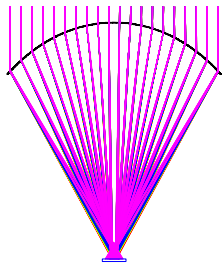
GEO Mission Array Comparisons

(from Murphy et al., IEEE PVSC, 5/02)

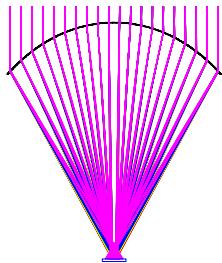
Specific Power (W/kg) for GEO Mission, 20 kW Array at EOL



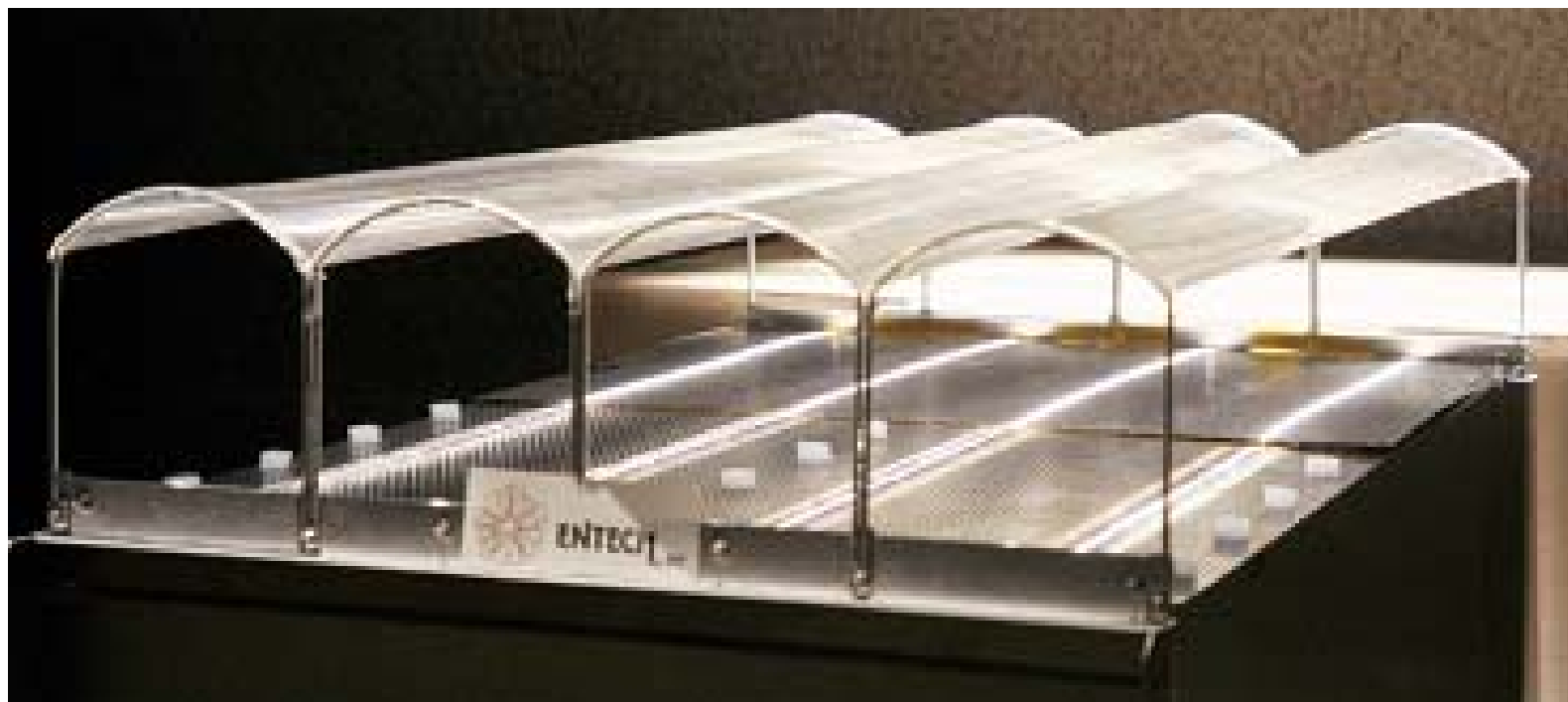
ABLE



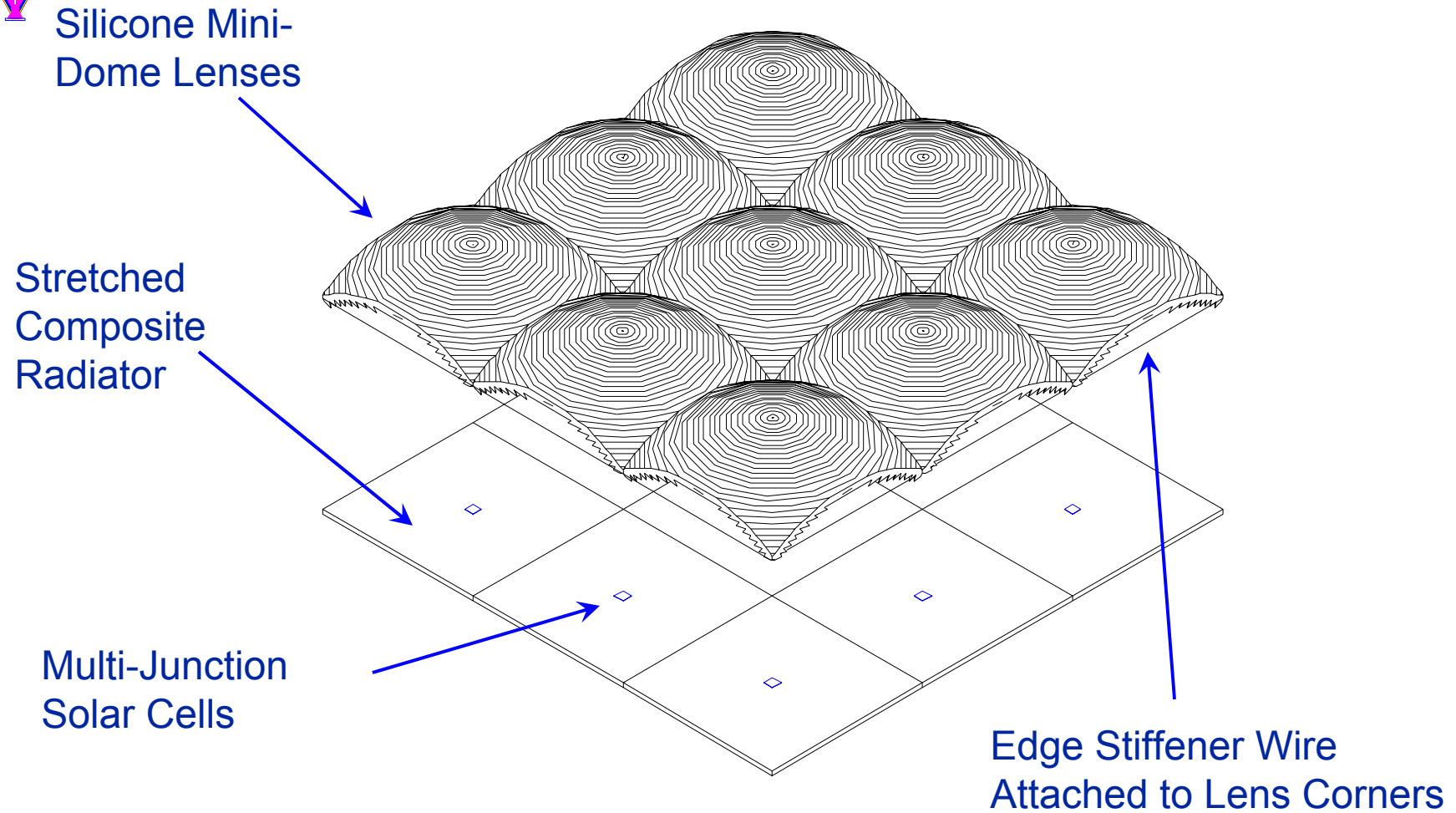
Concepts to Be Studied



Advanced Stretched Lens Linear Concentrator Blankets

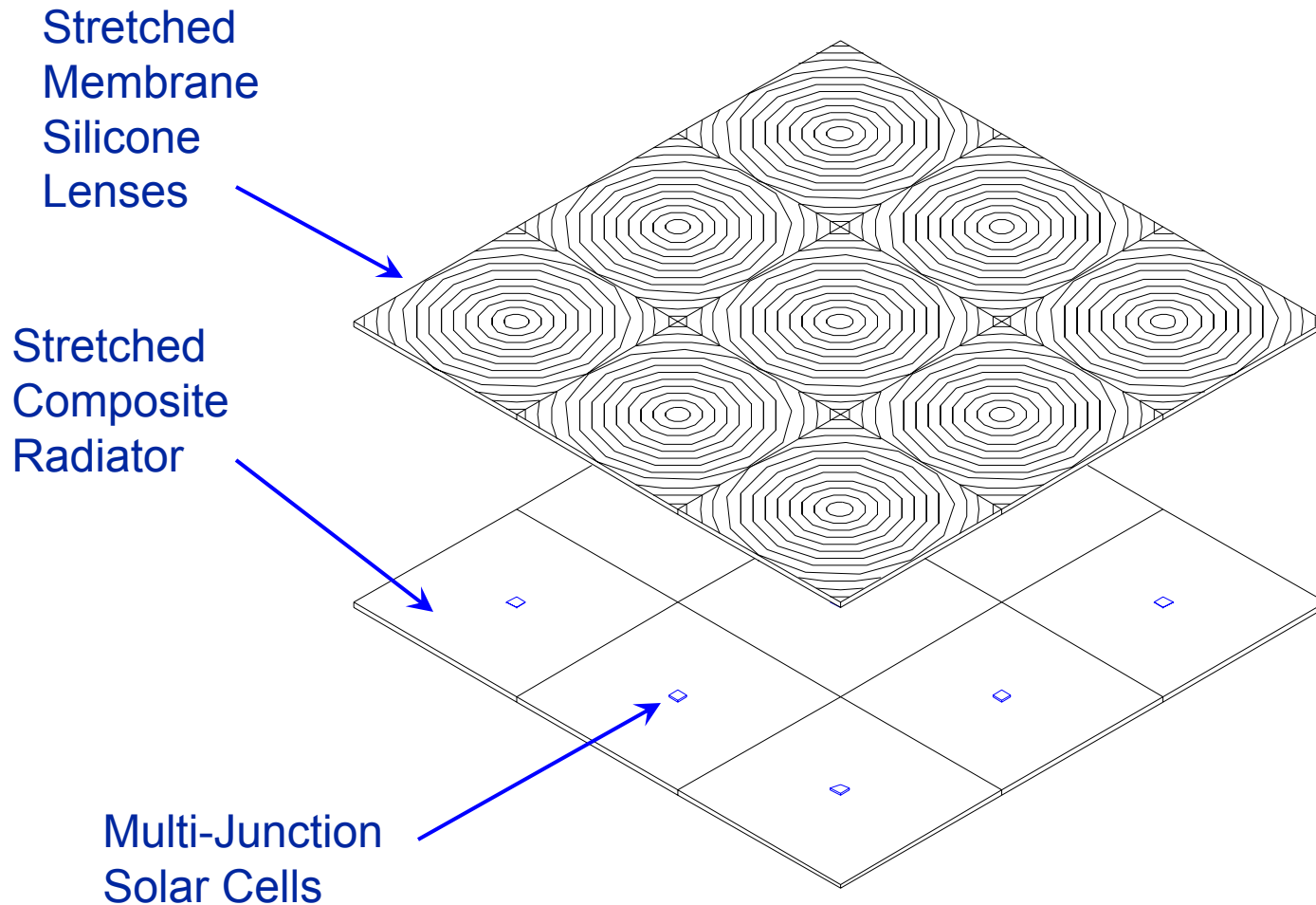


Conceptual Design of Advanced Mini-Dome Lens Array

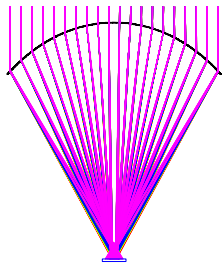


Flexible Domes Could Be Supported in Small Groups with Tensioned Edge Wires Providing Lens Layer Support

Conceptual Design of Flat Fresnel Lens Array

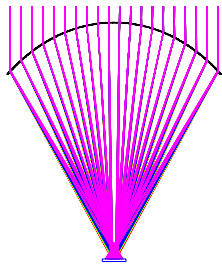


Optical Performance Will Suffer Compared to Arched or Dome Lens Designs, But Membrane Support Will Be Easier



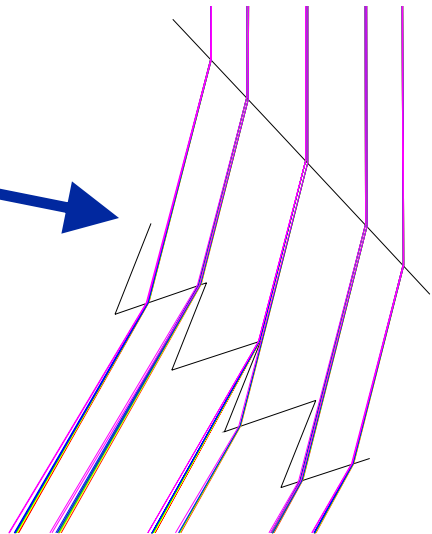
Optical Analysis Approach

Ray Trace for Color-Mixing Fresnel Lens



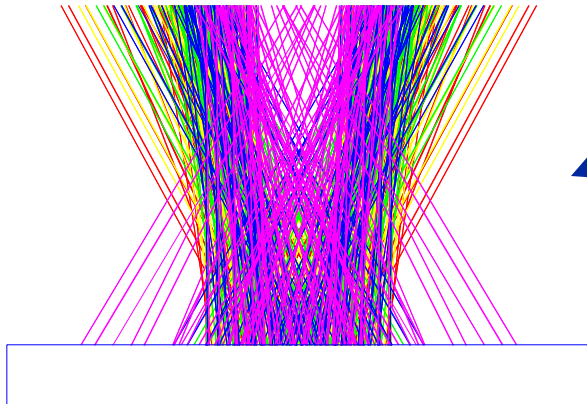
U.S. Patents
 4,069,812
 6,031,179
 6,075,200

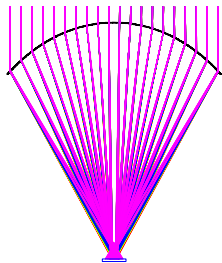
Receiver Close-Up



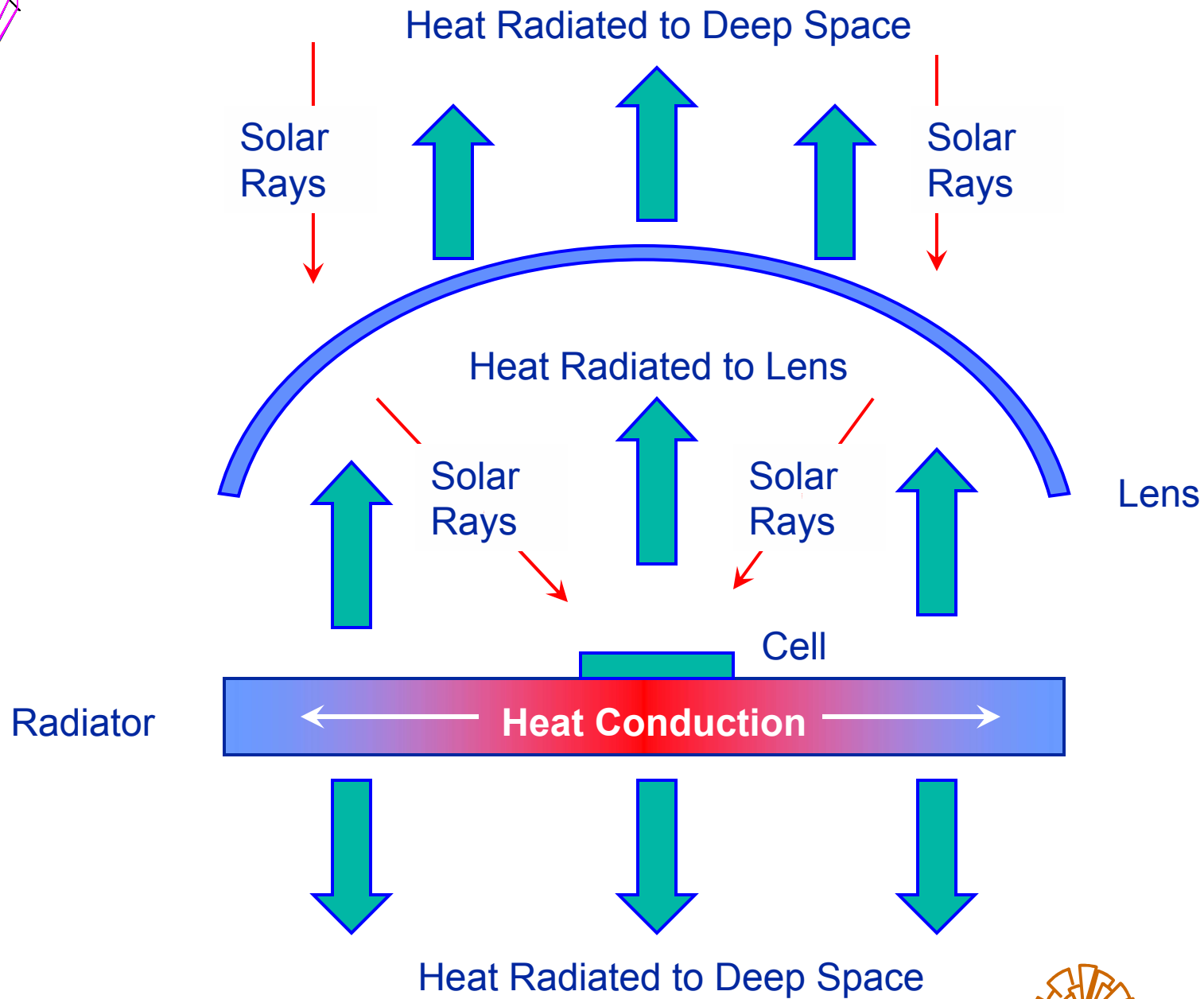
Lens Close-Up

**Every Other Prism Overlaps the
 “Blue” in Its Image with the
 “Red” in the Neighboring
 Prism’s Image**

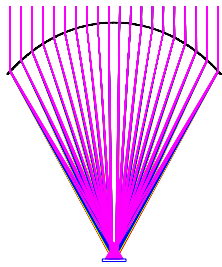




Thermal Analysis Approach



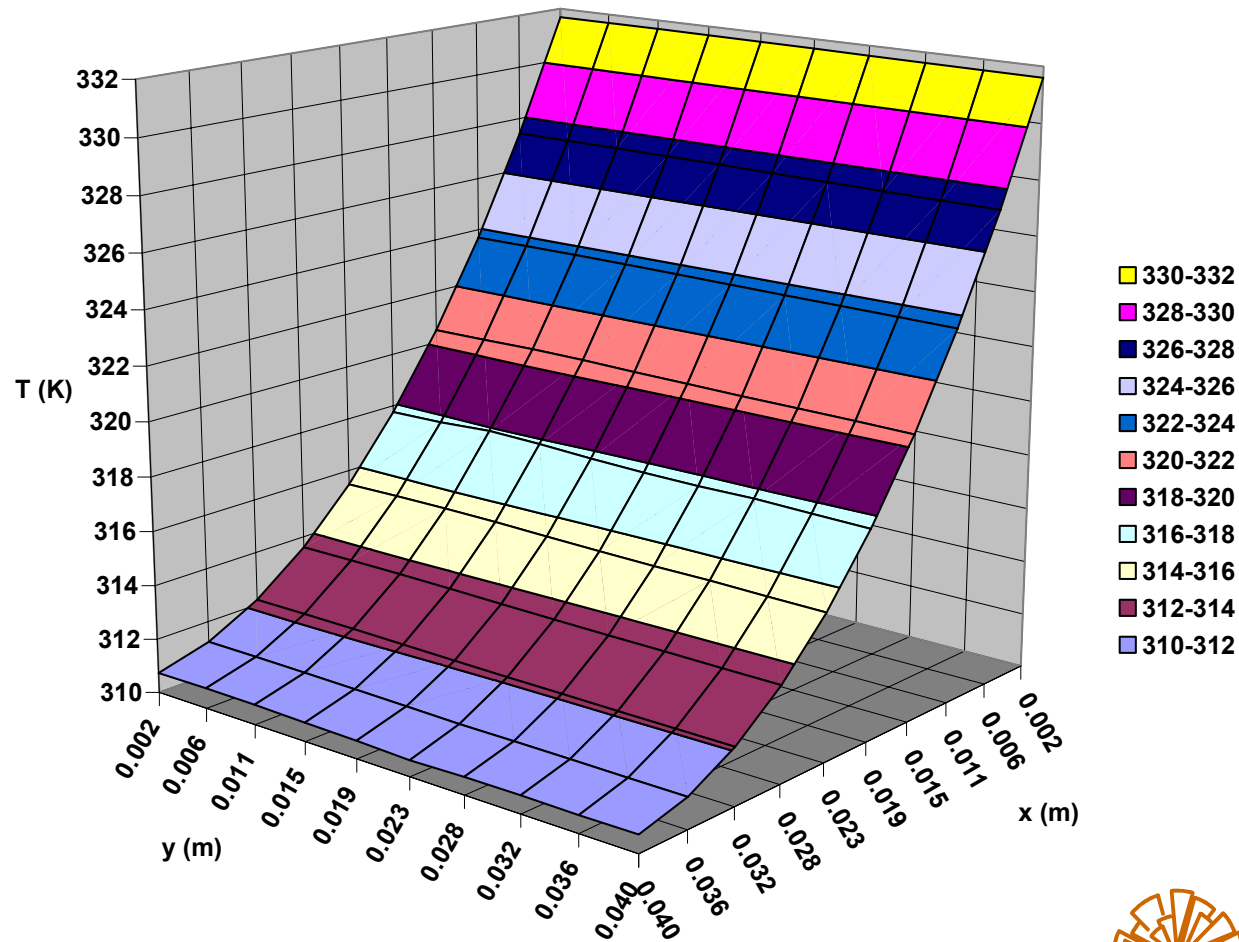
Thermal Analysis: Line Focus with 30% Cells



10X Linear Lens Radiator Thermal Analysis

8.5 cm Aperture Width, 30% Cell, 127 micron Radiator Thickness, GEO Orbit

Max Radiator Temperature Just Beneath Cell = 332K (59C)

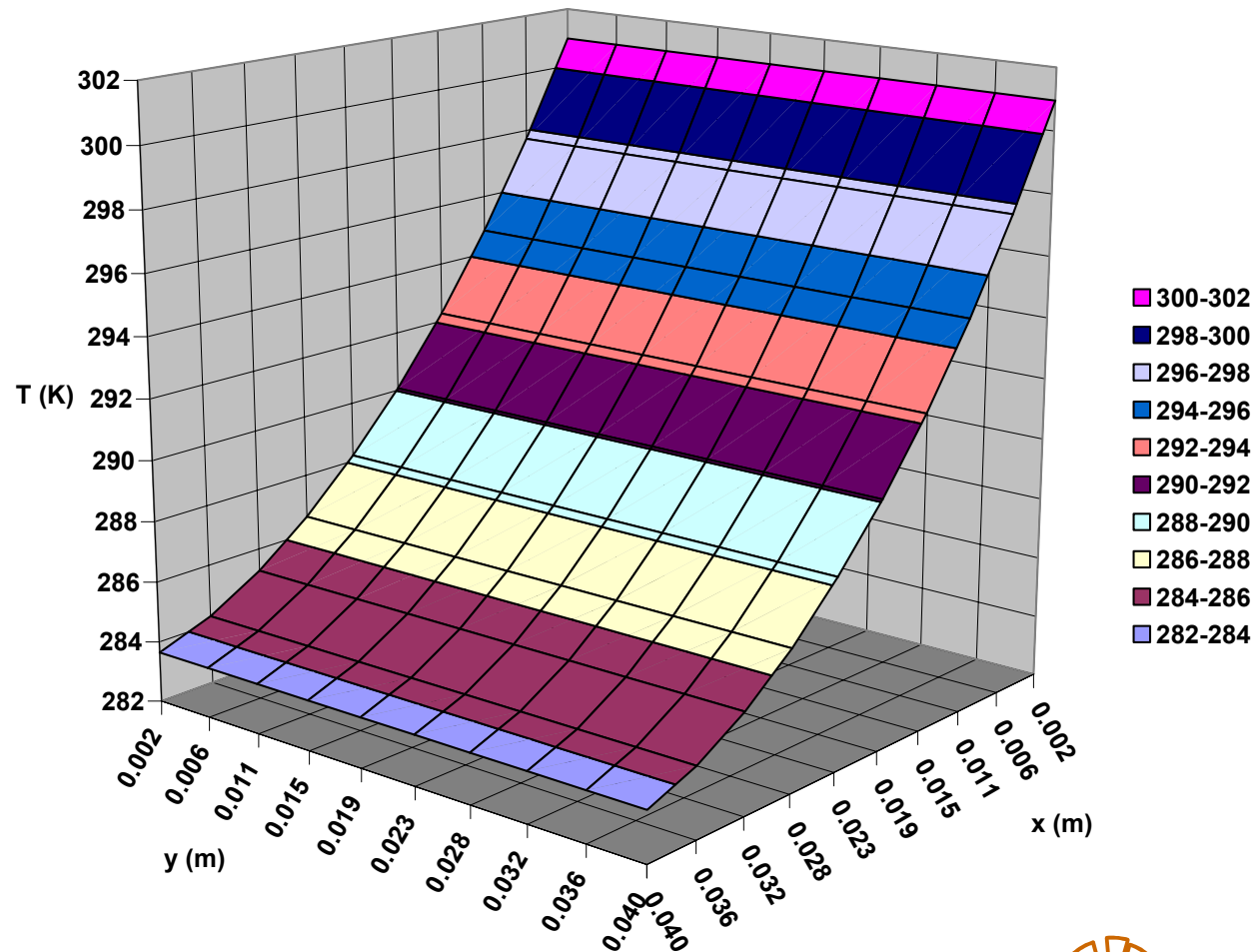


Thermal Analysis: Line Focus with 50% Cells

10X Linear Lens Radiator Thermal Analysis

8.5 cm Aperture Width, 50% Cell, 127 micron Radiator Thickness, GEO Orbit

Max Radiator Temperature Just Beneath Cell = 301K (28C)

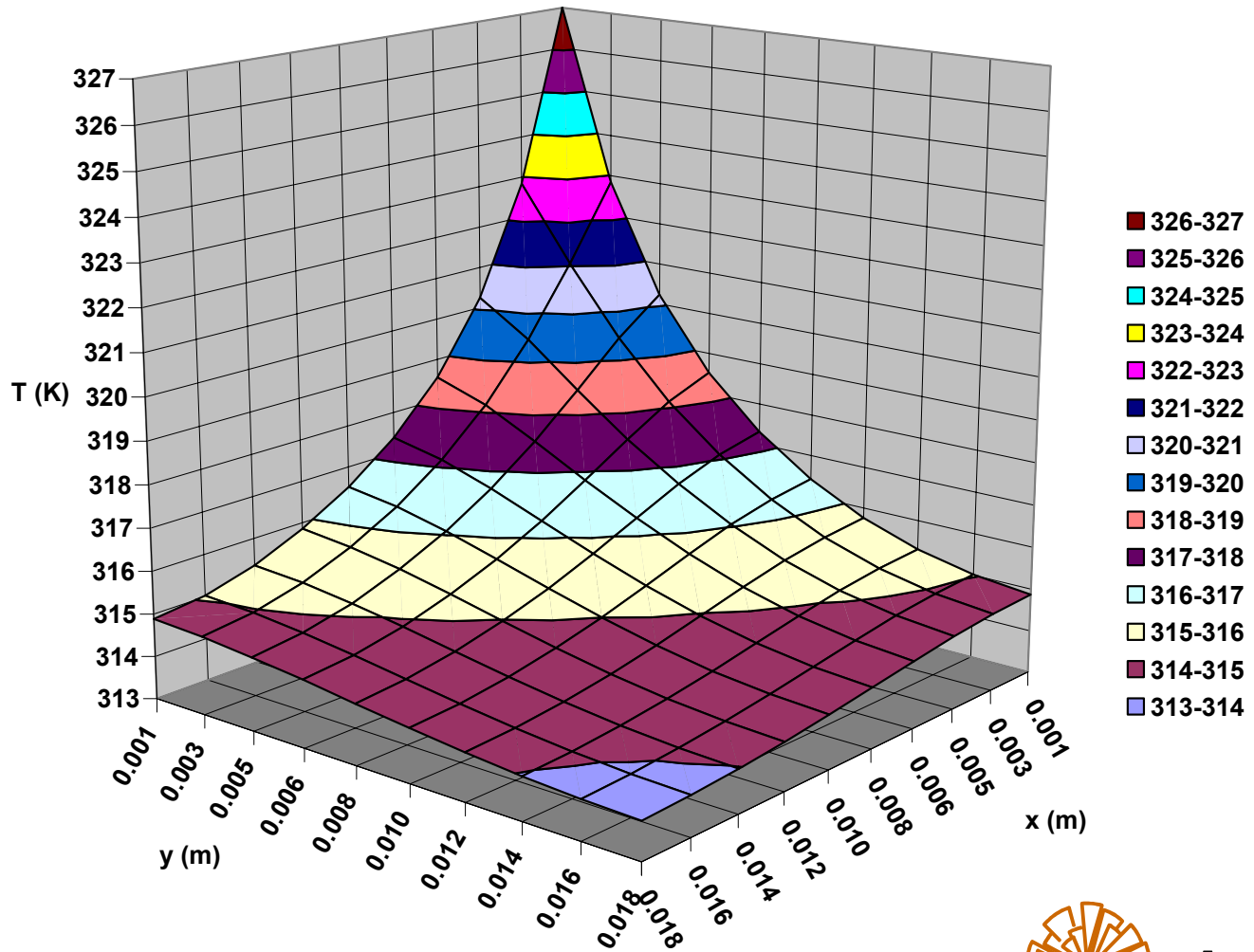


Thermal Analysis: Point Focus with 30% Cells

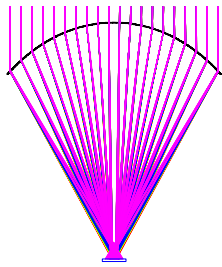
100X Mini-Dome Lens Radiator Thermal Analysis

3.7 cm x 3.7 cm Aperture Area, 30% Cell, 127 micron Radiator Thickness, GEO Orbit

Max Radiator Temperature Just Beneath Cell = 327K (54C)

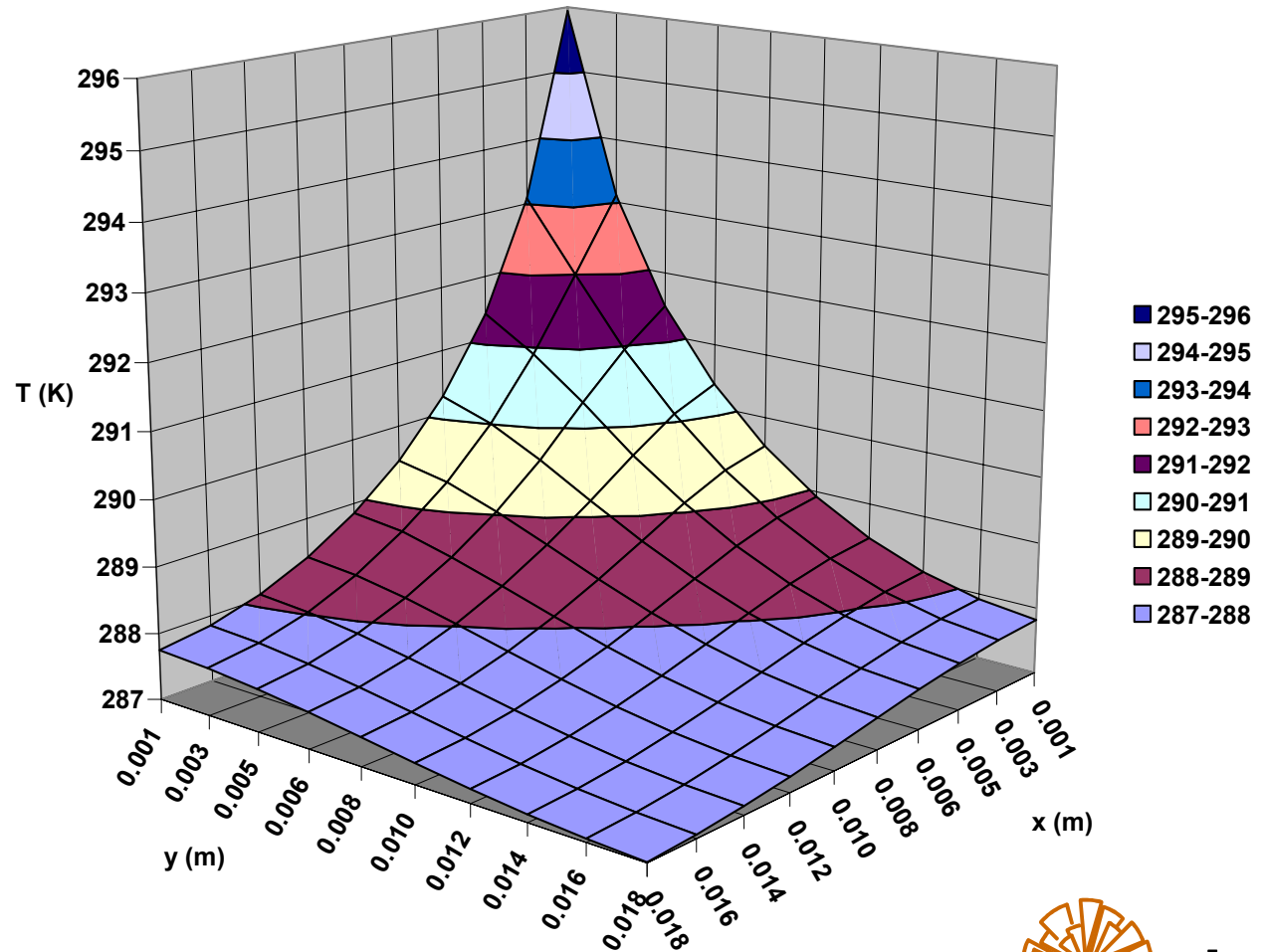


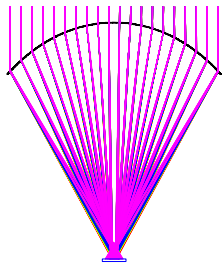
Thermal Analysis: Point Focus with 50% Cells



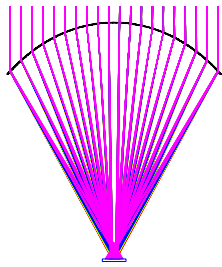
100X Mini-Dome Lens Radiator Thermal Analysis

3.7 cm x 3.7 cm Aperture Area, 50% Cell, 127 micron Radiator Thickness, GEO Orbit
Max Radiator Temperature Just Beneath Cell = 296K (23C)





Areal Mass Density Analysis Approach

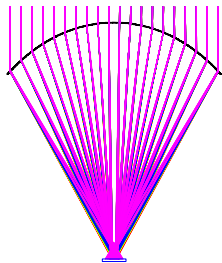


Example Areal Mass Density Calculation for Functional Concentrator Elements

Areal Mass Density of Active Lens, Radiator Sheet, and Photovoltaic Receiver Elements						
Geometric Concentration Ratio		100 X				
Major Subsystem	Element	Element Area per sq.m. Aperture	Thickness	Density	Mass/Aperture	Subtotals: Mass/Aperture
		(sq.m.)	(cm)	(g/cu.cm.)	(kg/sq.m.)	(kg/sq.m.)
Lens	Silicone Lensfilm	1.200	0.014	1.080	0.181	0.181
Radiator	Graphite/Cyanate Radiator	1.000	0.013	1.750	0.219	0.219
Receiver	Cell Prism Cover	0.010	0.025	1.080	0.003	0.016
	Germanium Cell Wafer	0.010	0.014	5.300	0.007	
	Conductive Silicone Adhesive	0.010	0.005	3.400	0.002	
	Flex Circuit Copper Traces	0.010	0.004	8.930	0.004	
	Flex Circuit Polyimide Film	0.010	0.005	1.500	0.001	
	Silicone Adhesive	0.010	0.003	1.080	0.000	
Total					0.417	0.417 kg/sq.m.

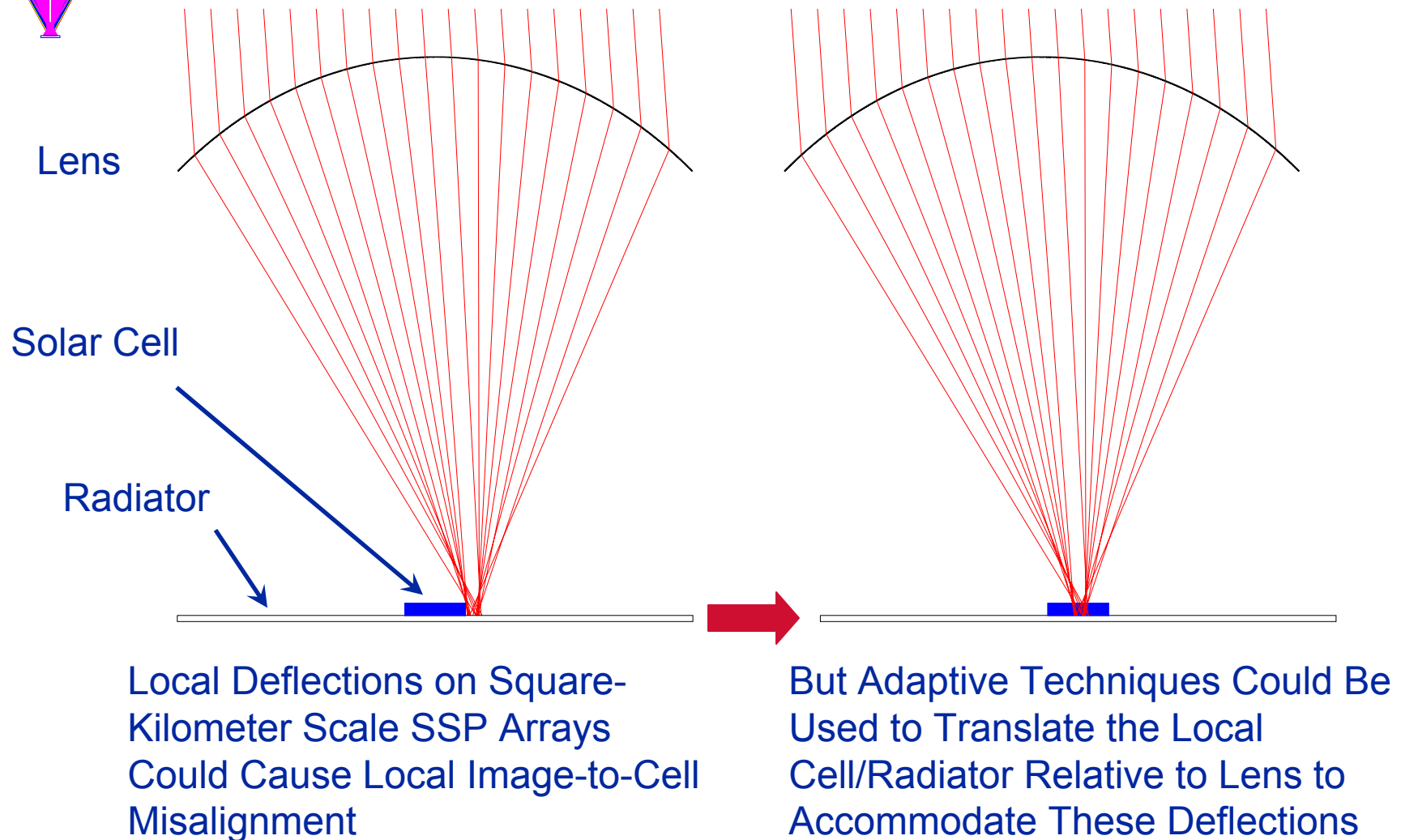
At SSP System Implementation	
Cell Efficiency	50%
Lens Efficiency	92%
Combined Efficiency	46%
Areal Power Density	629 W/sq.m.
Specific Power	1,509 W/kg

Summary of Parametric Results		
8.5 X	0.582 kg/sq.m.	1,081 W/kg
25 X	0.466 kg/sq.m.	1,349 W/kg
100 X	0.417 kg/sq.m.	1,509 W/kg

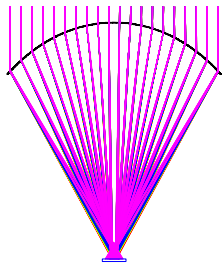


Basic Concept for Local Adaptive Techniques to Accommodate Local Deflections and Distortions Over Gigantic Concentrator Arrays

Adaptive Techniques Could Be Used to Accommodate Large SSP Deflections



A Flimsy Blanket Array with Large Deflections Can Work with a Concentrator Approach with Proper Adaptive Accommodation

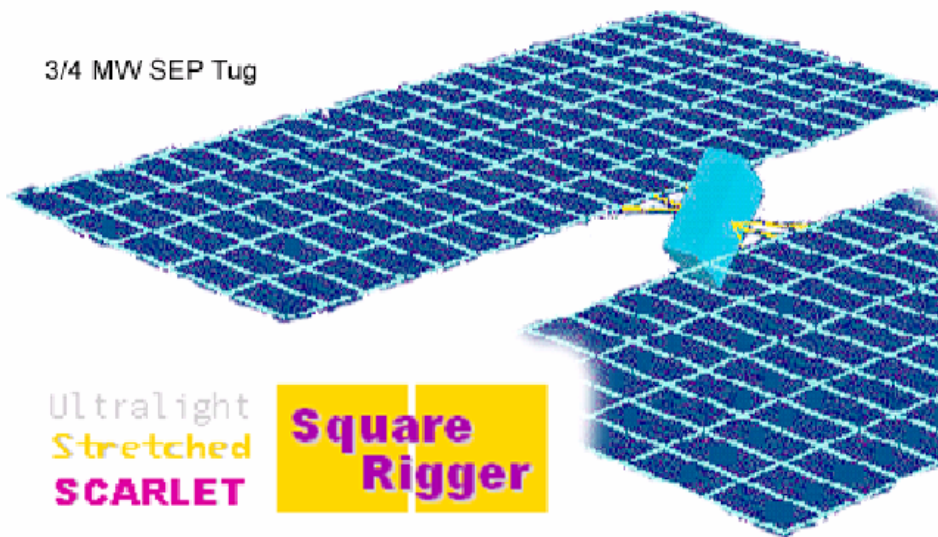


**The Ideal Ultra-Light, Scalable Platform for
Deployment and Support of Advanced
Concentrator Arrays:
ABLE's SquareRigger**

A Marriage Made in Heaven: ABLE's SquareRigger Platform and ENTECH's Stretched Lens Array

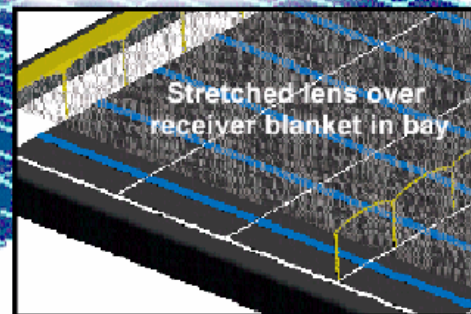
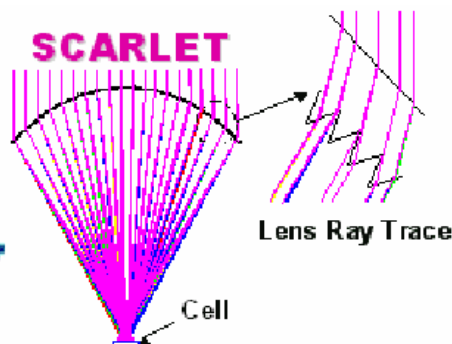


3/4 MW SEP Tug



Ultralight
Stretched
SCARLET

**Square
Rigger**



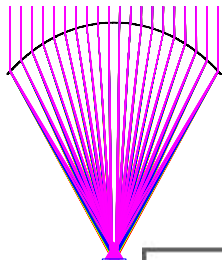
USS SquareRigger

Time Frame	< 5 Years	5-10 Years
Power Capability (kW)	100	1,000
BOL Specific Power (W/kg)	330	500
Stowed Power (kW/m ³)	80	120
Voltage	1,000	TBD

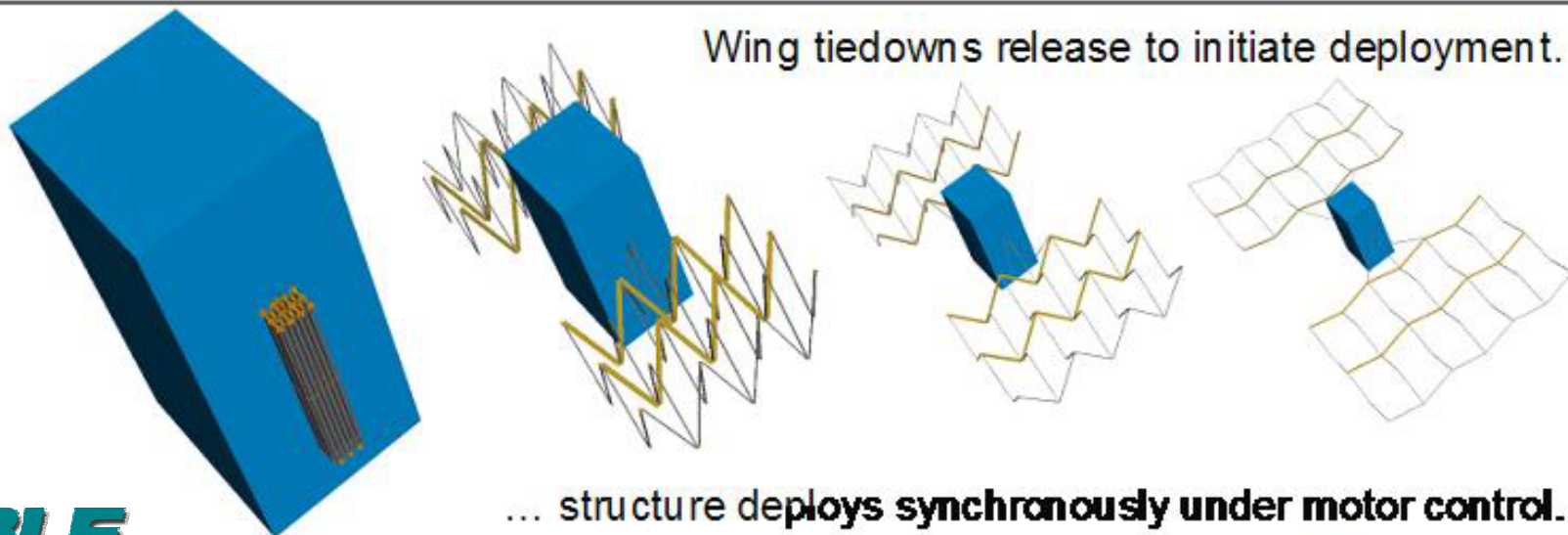
Applicable to: All NASA, DOD, and commercial spacecraft requiring high power, including:

- GEO communication satellites
- Interplanetary SEP spacecraft
- Space Solar Power
- SEP Orbital Tugs

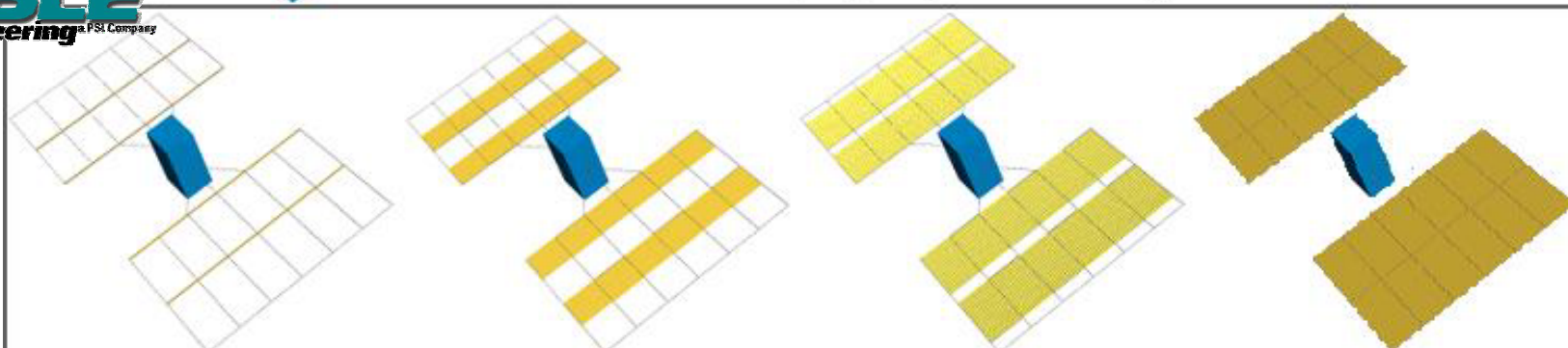
ABLE's SquareRigger Platform: Originally Developed for AFRL for Thin-Film Photovoltaics



Wing tiedowns release to initiate deployment...



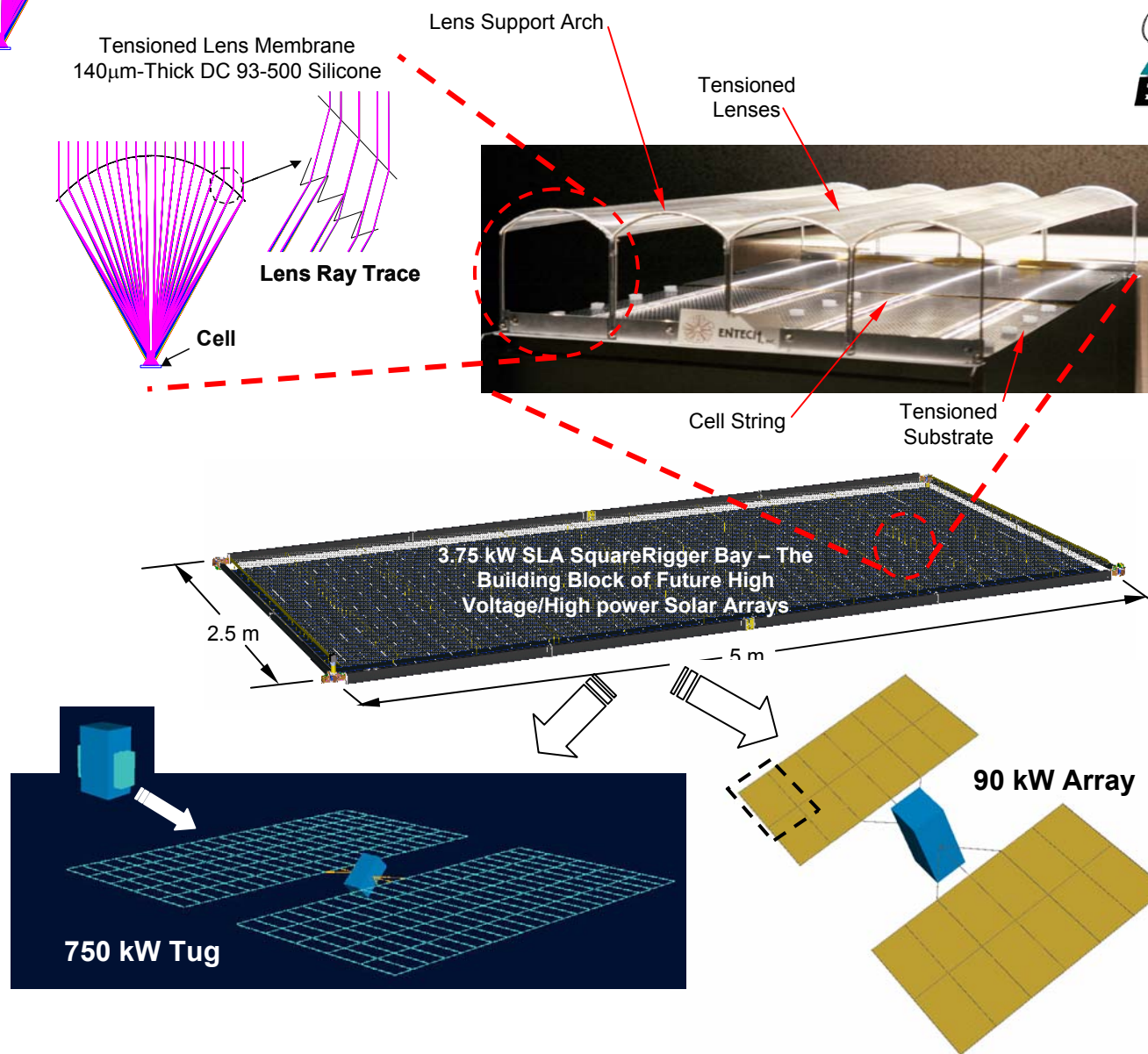
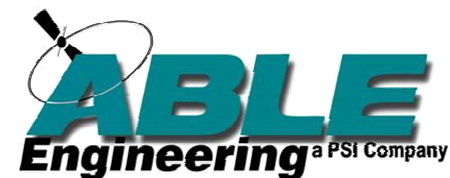
... structure deploys **synchronously under motor control**...



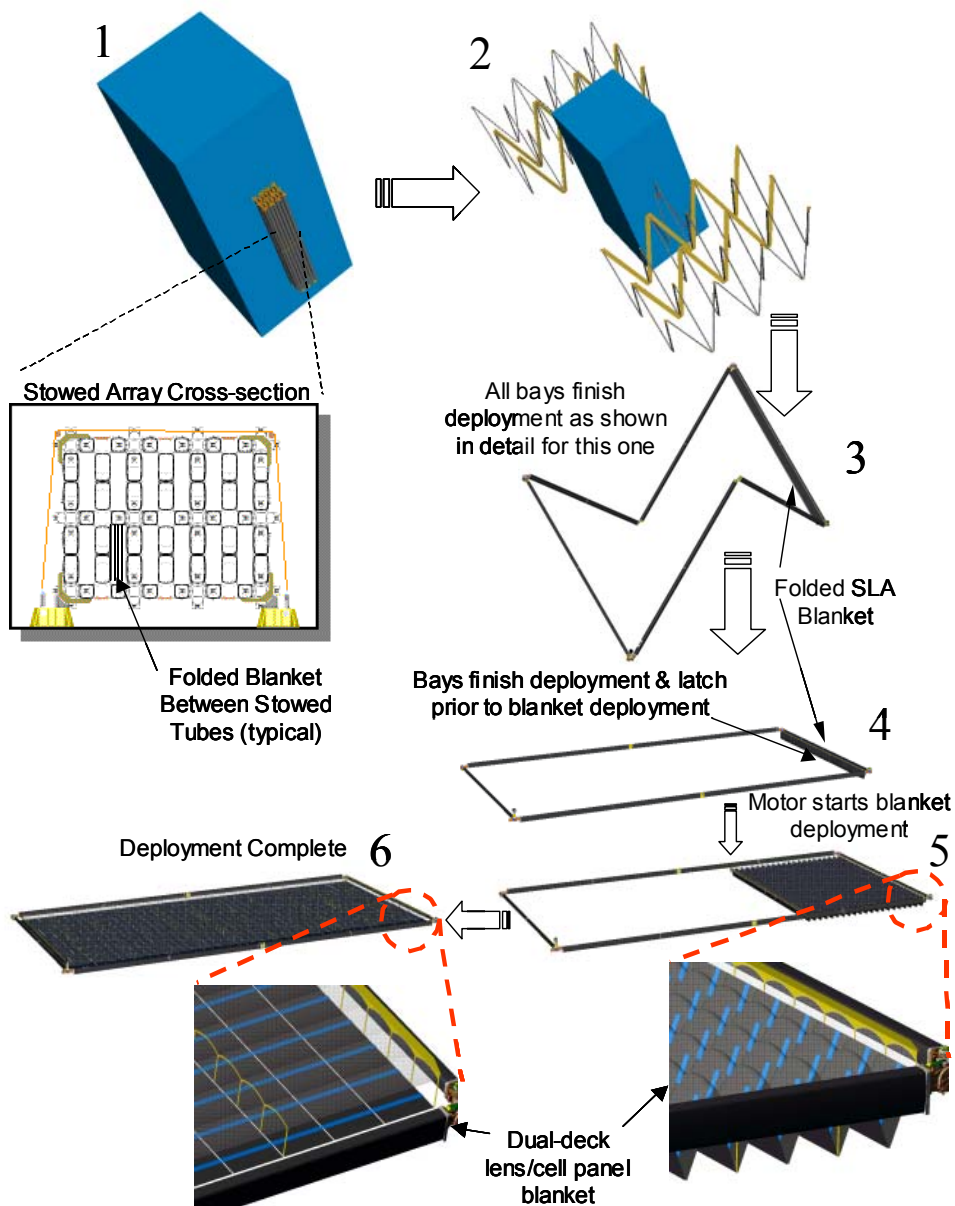
...once structure is latched, motors then work to raise photovoltaic blankets.



Ultralight Stretched SCARLET (USS) SquareRigger Concept

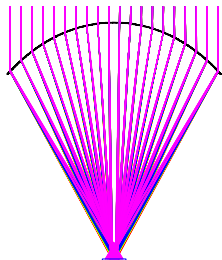


USS SquareRigger Deployment

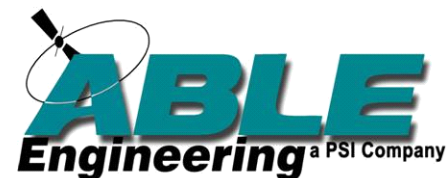


ABLE
Engineering a PSI Company

Full-Scale SquareRigger Bay Deployment

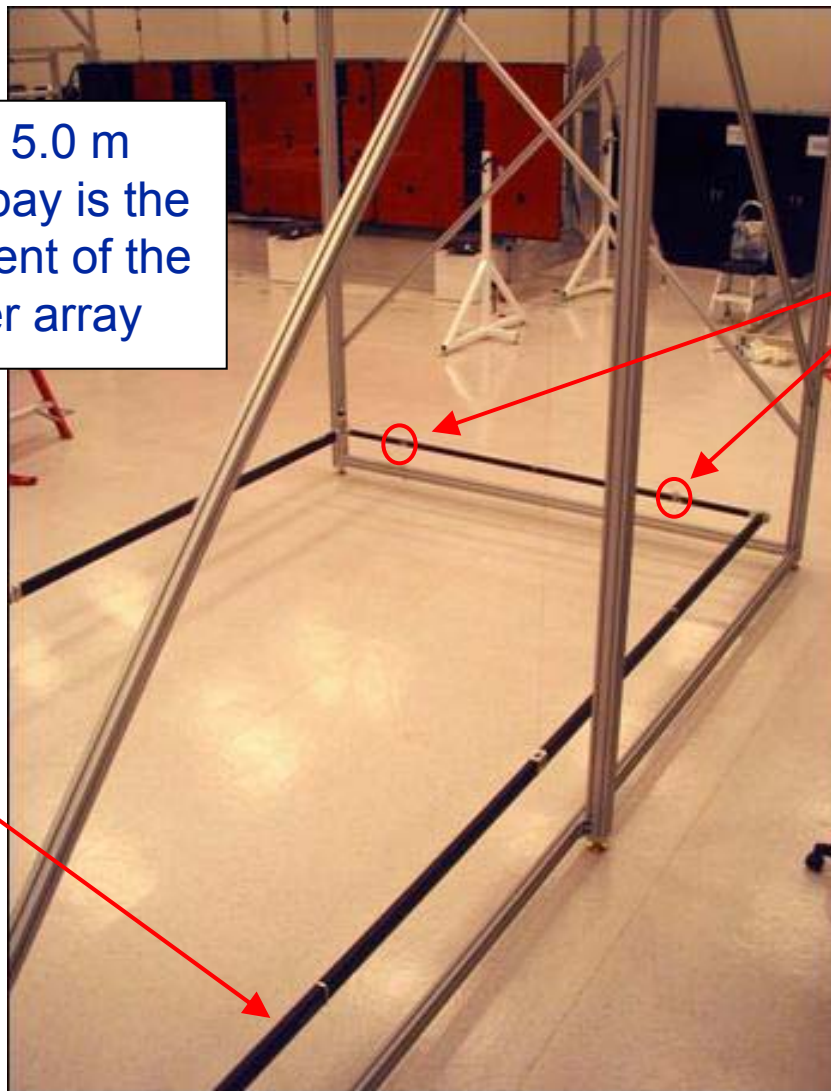


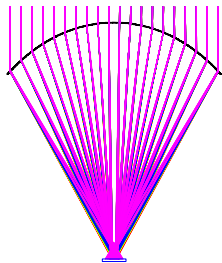
This 2.5 m x 5.0 m SquareRigger bay is the repeating element of the SquareRigger array



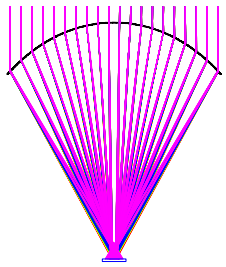
Ports for blanket deployment/tensioning cables

Low mass composite tube





Conclusions



Conclusions

- ◆ **Refractive Concentrators Using Multi-Junction Cells Represent the Most Efficient Option for Converting Sunlight to Electricity for Either Space or Ground Solar Power Applications**
- ◆ **Today's Demonstrated Lens/Cell Module Efficiencies Are 27% in Space and 30% on the Ground – Future Efficiencies at SSP Implementation (e.g., 2022) Should Nearly Double Based on Cell Efficiency Trends**
- ◆ **Compared to Thin Film Arrays, Refractive Concentrator Arrays Will Be 2-4X More Efficient, 2-4X Smaller in Size, and Equal or Better in Terms of W/kg and \$/W at the System Level**
- ◆ **Work Should Continue on Advanced Concentrator Arrays for SSP, Including Advanced Ultra-Light Concepts and Simple Means to Accommodate Large Deflections on Square-Kilometer Size Arrays, Using Techniques from Adaptive Optics, Smart Materials, Shape Memory Alloys, etc.**